

SmartAnswer – Investigations of stream-wise Rod Vortex Generators effect on noise generated by boundary layer separation on blades.

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Smart Mitigation of flow-induced Acoustic Radiation and Transmission for reduced Aircraft, surface traNSport, Workplaces and wind enERgy noise



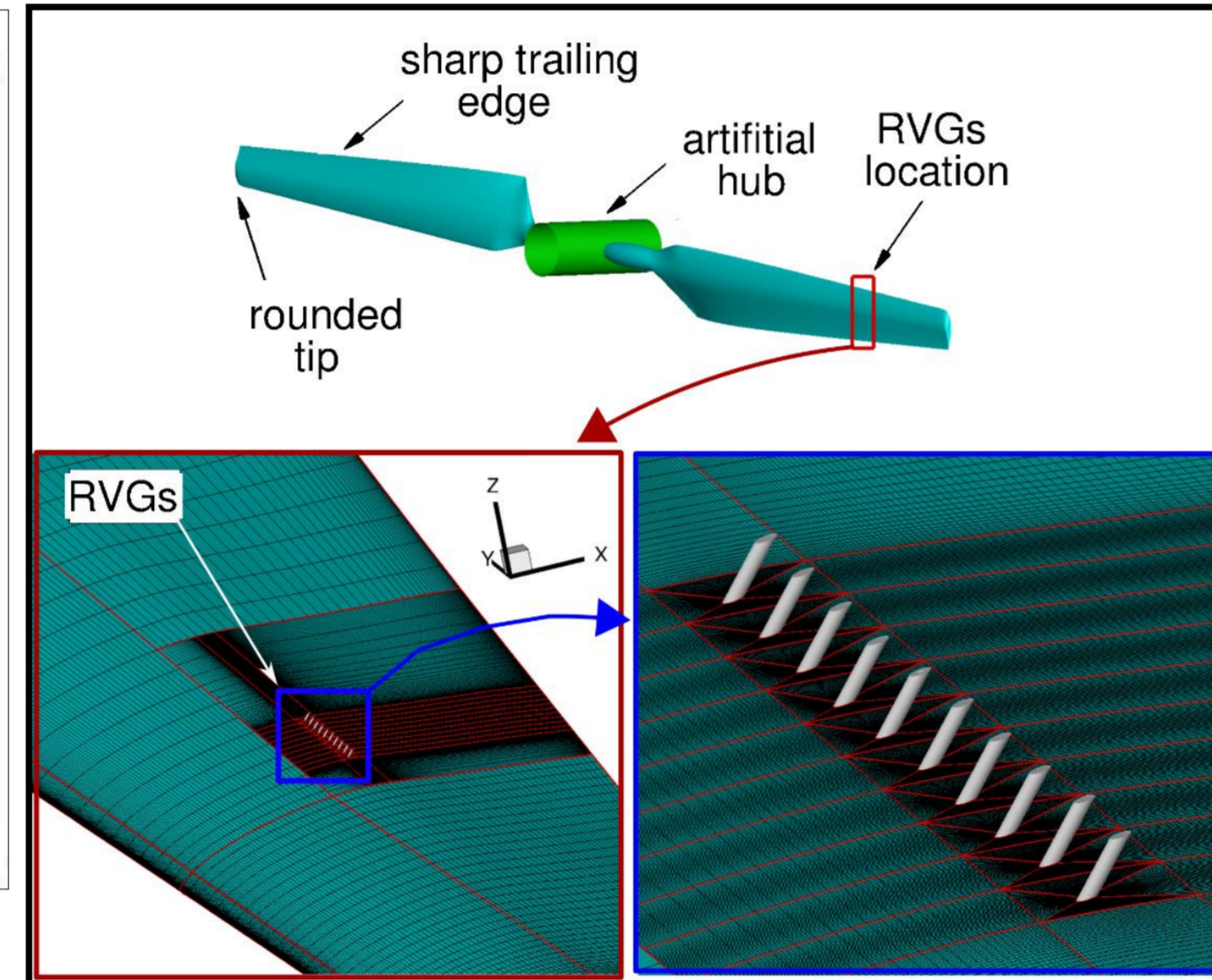
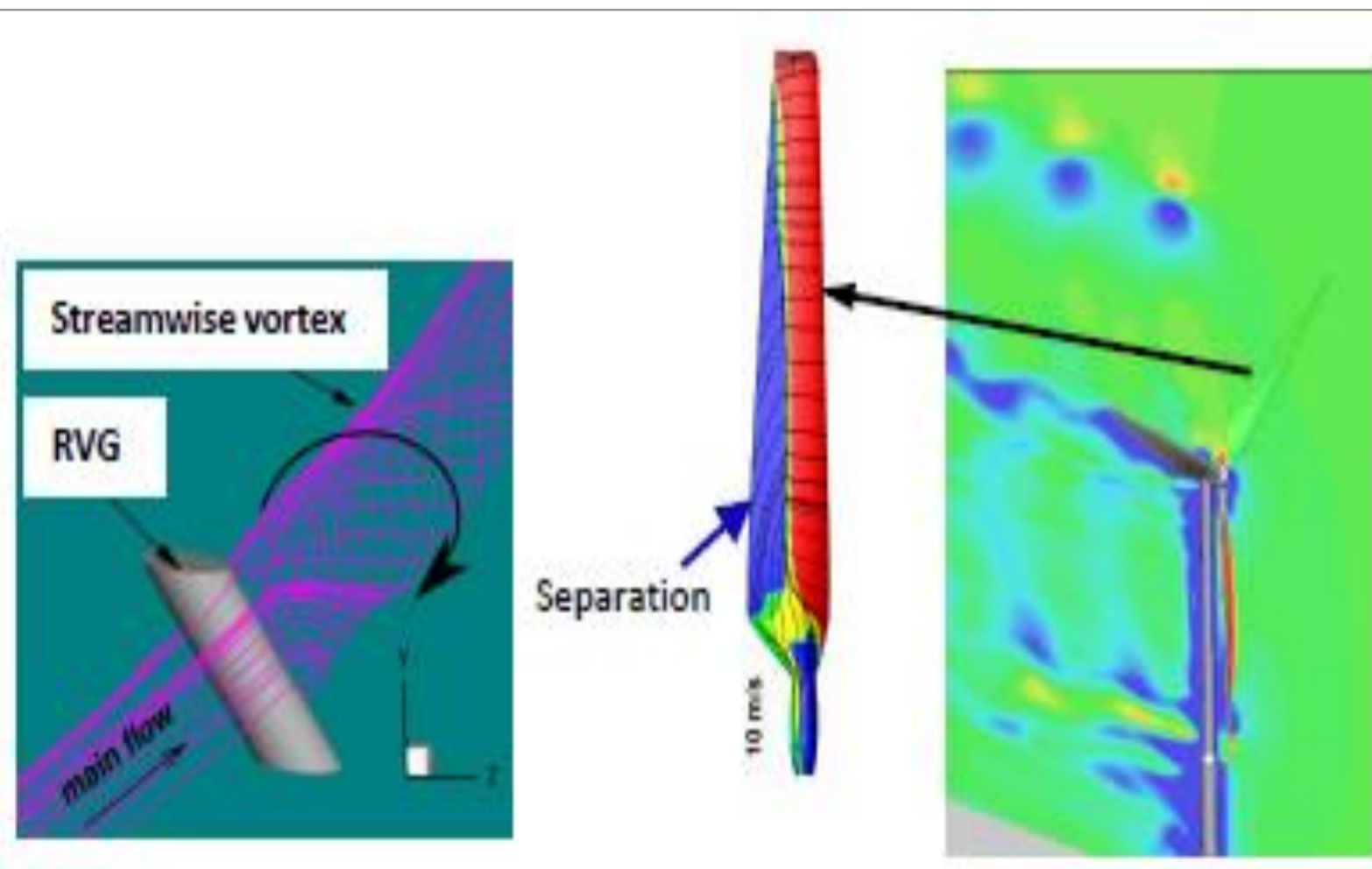
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Motivation

- One of the challenges for **wind energy** is to overcome the **noise** complaint of the residents living near wind farms.
- **Trailing edge noise** has been studied intensively for **design** cases.
- **Flow separation** occurs in higher wind speed cases leading to **decreased performance** of wind turbines in **off-design** conditions.



Flow visualization on wind turbine blade with RVGs [1]

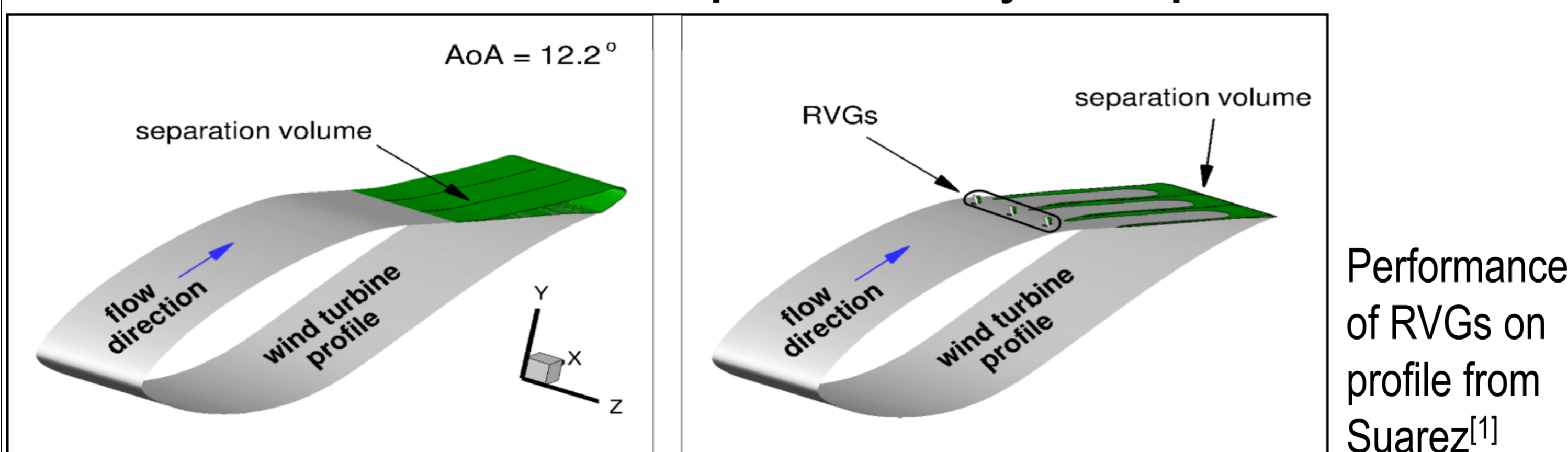
- **Rod Vortex Generators (RVGs)** are investigated to **reduce the flow separation**.
- **RVGs increase aerodynamic performance** of wind turbine and helicopter rotors.
- What is their **acoustic** impact? Increase or decrease noise levels?

Main Objectives

- To identify **noise sources** in the flow field for **wind turbine blades** in the **boundary layer separation** cases.
- To investigate the effect of **RVGs on noise** generated by rotor blade.

Existing Background

- Numerical investigations of **RVGs installed on helicopter rotor** and **NREL wind turbine rotor** show **improved aerodynamic performance**.



Performance of RVGs on profile from Suarez [1]

Methodology

- **Ffowcs Williams- Hawkings (FW-H)** acoustic analogy describes **acoustic pressure** as a sum of elementary acoustic sources such as the **monopole, dipole and quadrupole**.
- **Aero-acoustic code development** in **Tecplot** using macro scripting based on **linear integral Farassat Formulations of FW-H** in **time domain**.
- Acoustic investigations of **rotating bodies** such as **helicopter rotor, wind turbine**.
- **Experimental acoustic and flow (PIV) measurements** for wind turbine profile **DU96-W-180 reference case** and **RVGs case** (designed to control separation) in test section at **TU Delft**.

Theory: Integral solution of FW-H by Farassat as Formulation 1A [2]

$$4\pi p'_T(x, t) = \int_{f=0} \left[\frac{\rho_0 \dot{v}_n}{r(1-M_r)^2} + \frac{\rho_0 v_n \hat{r}_i \dot{M}_i}{r(1-M_r)^3} \right]_{ret} dS + \int_{f=0} \left[\frac{\rho_0 c v_n (M_r - M^2)}{r^2(1-M_r)^3} \right]_{ret} dS$$

Thickness near-field term Thickness far-field term

$$4\pi p'_L(x, t) = \int_{f=0} \left[\frac{\dot{p} \cos \theta}{cr(1-M_r)^2} + \frac{\hat{r}_i \dot{M}_i p \cos \theta}{cr(1-M_r)^3} \right]_{ret} dS + \int_{f=0} \left[\frac{p(\cos \theta - M_i n_i)}{r^2(1-M_r)^2} + \frac{(M_r - M^2)p \cos \theta}{r^2(1-M_r)^3} \right]_{ret} dS$$

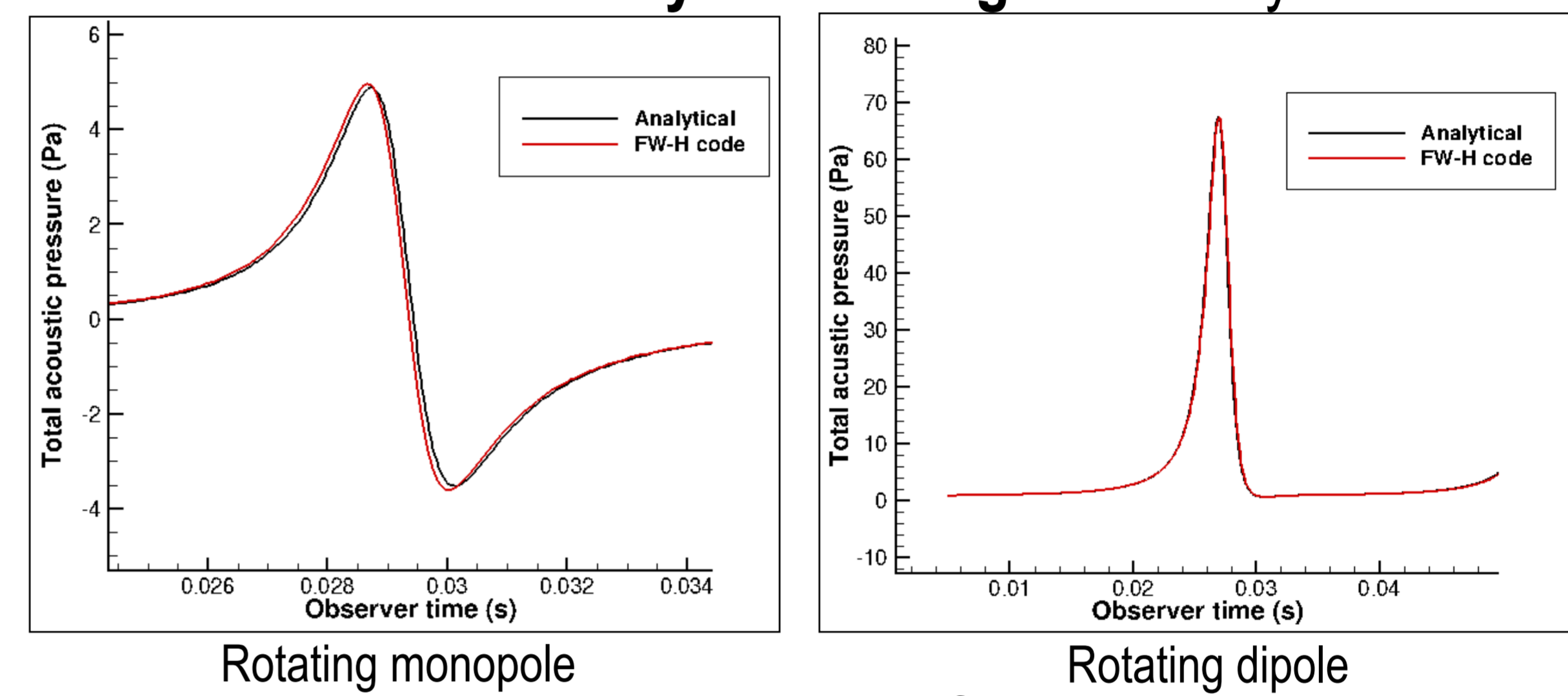
Loading near-field term Loading far-field term

Analysis:

- **Validation**
 - **Stationary : Monopole/Thickness and Dipole/loading**
 - **Rotating : Monopole, Dipole, source - sink pair**
- **Helicopter rotor case** – experimental and other numerical results available (**thickness term dominant**).
- **Wind turbine blade** with and without **RVGs** (**loading term dominant**).

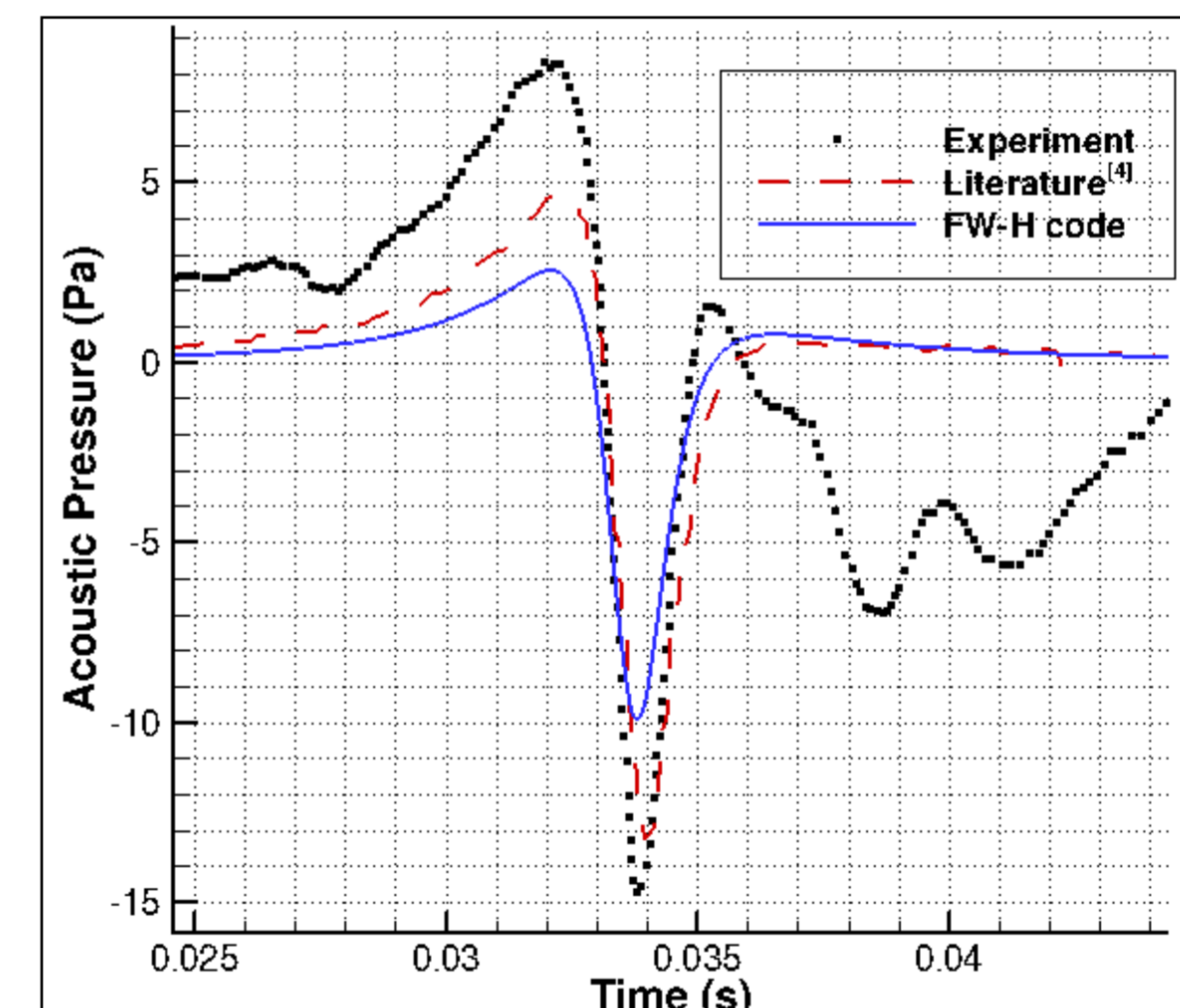
Results

- **Validation of elementary sources against analytical solutions [3]**



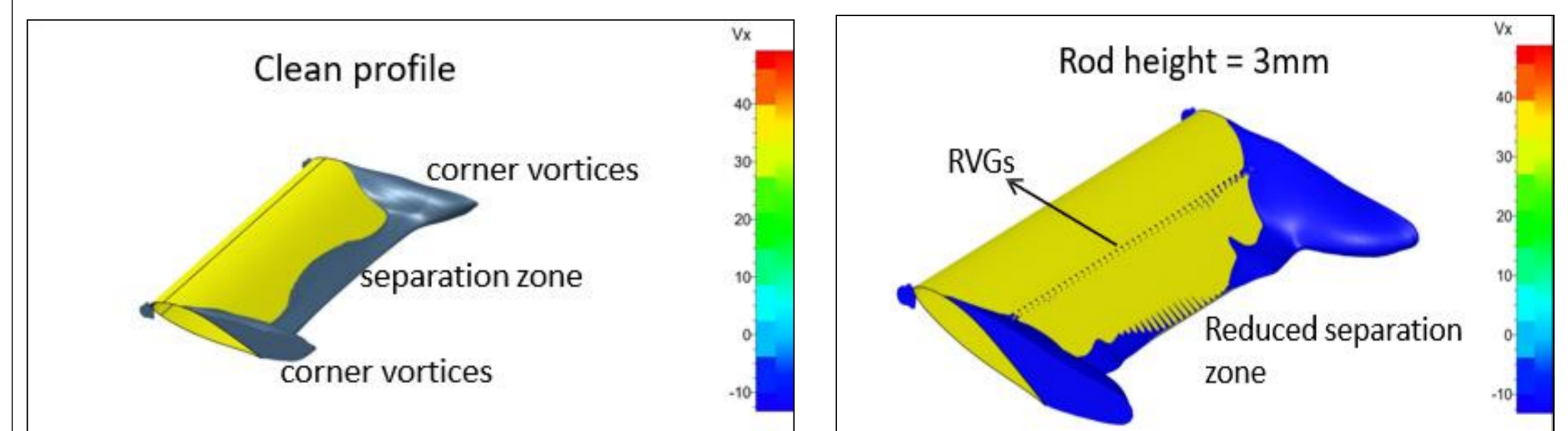
Good agreement with analytical solutions

- **Helicopter results:** Acoustic signal from code against experimental data and numerical acoustic model for 1/7 scale model helicopter by **Sargent [4]** at **M_T = 0.661**, hover mode.



Good trends. Under-prediction of peak amplitude under analysis.

- **Design of RVGs for DU96-W-180 profile** for an anechoic vertical open-jet wind tunnel at **TU Delft [5]**



DU96-W-180 profile, full 3D simulation at flow velocity = 30 m/s, AoA = 13°

Future Work

- **Steady and unsteady pressure data** of **NREL Phase VI rotor** is under acoustic analysis using the developed **aero-acoustic code**.
- **Secondment at TU Delft** : **experimental acoustic and flow measurements** for **DU96-W-180** with/without RVGs.

References

- [1] Suarez, et al, *Wind Energy*, 21:1202-1215, 2018.
- [2] Farassat, F., *NASA TM-214853*, 2007.
- [3] Rienstra, S. W. et al, *IWDE92-06*, 2001.
- [4] Sargent, D. C. et al, *Jou. of Aircraft*, 2014.
- [5] Carpio, A. R. et al, *Jou. of Sound and Vibration*, 2018.



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